

Strömung, Übung 7

Oppg. 1a)

(I) $\frac{du}{dx} + \frac{dv}{dy} = 0 \rightarrow \frac{d\rho}{dt} = 0$, stationäre Strömung, eller inkompressibel

(II) $u \frac{du}{dx} + v \frac{du}{dy} = \nu \left(\frac{d^2 u}{dx^2} + \frac{d^2 v}{dy^2} \right)$
 $\rightarrow \frac{du}{dx} = \frac{dv}{dy} = 0 \rightarrow 2D$ Strömung
 $\nabla p = 0$, $\nabla \cdot \mathbf{v} = 0$, $\frac{d}{dt}(\rho u)$

1b) Skalierung med

$$x^* = \frac{x}{L}, \quad y^* = \frac{y}{\delta}$$

$$u^* = \frac{u}{U_\infty}, \quad v^* = \frac{v}{V_\infty}$$

I) $\frac{U_\infty}{L} \frac{d^2 u^*}{dx^{*2}} + \frac{V_\infty}{\delta} \frac{d^2 v^*}{dy^{*2}}$

$$\frac{du^*}{dx^*} + \frac{LV_\infty}{U_\infty \delta} \frac{dv^*}{dy^*} = 0 \rightarrow V_\infty = \frac{U_\infty \delta}{L}$$

II) $U_\infty u^* \frac{U_\infty}{L} \frac{du^*}{dx^*} + \frac{U_\infty \delta}{L} v^* \frac{U_\infty}{\delta} \frac{dv^*}{dy^*} = \nu \left(\frac{U_\infty}{L^2} \frac{d^2 u^*}{dx^{*2}} + \frac{U_\infty \delta}{\delta^2} \frac{d^2 v^*}{dy^{*2}} \right)$

$$\frac{U_\infty^2}{L} \left(u^* \frac{du^*}{dx^*} + v^* \frac{dv^*}{dy^*} \right) = \frac{\nu U_\infty}{\delta^2} \left(\frac{\delta^2}{L^2} \frac{d^2 u^*}{dx^{*2}} + \frac{d^2 v^*}{dy^{*2}} \right)$$

$$u^* \frac{du^*}{dx^*} + v^* \frac{dv^*}{dy^*} = \frac{\nu L}{U_\infty \delta^2} \frac{d^2 u^*}{dx^{*2}}$$

$$\approx 1 \Rightarrow \delta = \sqrt{\frac{\nu L}{U_\infty}}$$

$$\delta = \sqrt{\frac{\nu L}{U_\infty}} = L \sqrt{\frac{\nu}{U_\infty L}} \rightarrow \frac{\delta}{L} = \sqrt{\frac{1}{Re_L}}$$

$$\delta \ll L \rightarrow \frac{1}{Re_L} \ll 1 \rightarrow Re_L \gg 1$$

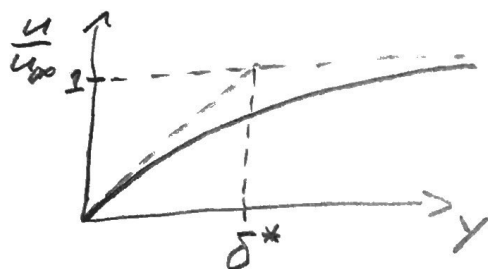
Oppg. 2a)

boundary layer thickness:

$$u(y=\delta) = 0,99 u_\infty$$

Punktet hvor hastigheten er 99% av
Free-Flow hastighet

Film thickness



tykkelsen til δ -iktet
der som hastigheten hadde
vært lineært med

$$\frac{du}{dy} = \left. \frac{du}{dy} \right|_{y=0}$$

Displacement thickness

$$\delta^{**} = \int_0^\infty \left(1 - \frac{u}{u_\infty}\right) dy$$

Er relatert til aerodynamiske og trykkløst

2b) $C_D = \frac{1}{\frac{1}{2} \rho u_\infty^2} \left(\frac{F_D}{A_F^*} \right)$ ← relatert til motstand
når fluid strømmer rundt
noe

$C_F = \frac{1}{\frac{1}{2} \rho u_\infty^2} \tau_w$ ← relatert til skjærekraft
på en flate

2c) $\delta = 4,92 \sqrt{\frac{\nu x}{u_\infty}}$

$C_F = 0,664 \sqrt{\frac{\nu}{u_\infty x}}$

$\tau_w = \frac{1}{2} \rho u_\infty^2 C_F$

i) $\delta = 1,5 \text{ mm}$

ii) $\delta = 1,5 \text{ cm}$

i) $C_F = 2,02 \cdot 10^{-3}$

ii) $C_F = 2,02 \cdot 10^{-2}$

i) $\tau_w = 0,175 \text{ Pa}$

ii) $\tau_w = 1,75 \text{ Pa}$

i) $\mu = 10^{-2} \text{ Pa s}$

ii) $\mu = 0,1 \text{ Pa s}$

Oppg 2d)

$$F_w = \int \tau_w dA = 0,5 \int_0^{0,5} \tau_w dx$$
$$= 0,5 \int_0^{0,5} \frac{1}{2} g u_0^2 \cdot 0,664 \sqrt{\frac{\nu}{u_0 x}} dx$$

$$= \frac{0,664}{4} g u_0^2 \sqrt{\frac{\nu}{u_0}} \left[2\sqrt{x} \right]_0^{0,5}$$

$$F_w (\mu = 0,1 \text{ Pas}) = 0,6 \text{ N}$$

$$F_w (\mu = 10^{-3} \text{ Pas}) = 61,7 \text{ mN}$$

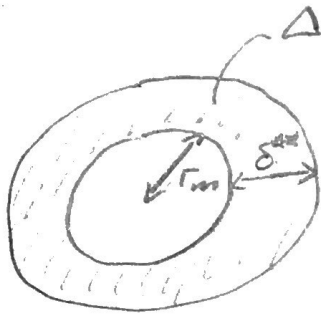
Oppg. 3a)

$$(I) R = \frac{g_3 u_m \Delta}{g_m u_m \pi \Gamma_m^2}$$

$$(II) \delta^{**} = 1,72 \sqrt{\frac{\nu x}{u_m}}$$

For at resultatet for plate skal være gyldige må $\delta^{**} \ll \Gamma_m$

3b)



$$\Delta = \pi (\Gamma_m + \delta^{**})^2 - \pi \Gamma_m^2$$

$$= \pi ((\Gamma_m + \delta^{**})^2 - \Gamma_m^2)$$

$$\Delta = \pi (2\Gamma_m \delta^{**} + (\delta^{**})^2) = \pi \delta^{**} (2\Gamma_m + \delta^{**})$$

$$\approx 2\pi \Gamma_m \delta^{**}$$

$$R = \frac{g_3 u_m 2\pi \Gamma_m \delta^{**}}{g_m u_m \pi \Gamma_m^2}$$

$$R = \frac{2g_3 \delta^{**}}{g_m \Gamma_m} = \begin{cases} \text{Case 1: } 5 \cdot 10^{-4} \\ \text{Case 2: } 3,76 \cdot 10^{-5} \end{cases}$$

NB: case 1 gir $\delta^{**} = 1,72 \cdot 10^{-2} > \Gamma_m$

→ Ligningen holder ikke

Grafisk:

$$\begin{cases} \text{Case 1: } \frac{\nu x}{u_m \Gamma_m^2} = 11 \rightarrow \frac{\Delta}{\pi \Gamma_m^2} \approx 12 \rightarrow R = 5,7 \cdot 10^{-4} \\ \text{Case 2: } \frac{\nu x}{u_m \Gamma_m^2} = 5 \cdot 10^{-2} \rightarrow \frac{\Delta}{\pi \Gamma_m^2} \approx 1 \rightarrow R = 4,7 \cdot 10^{-5} \end{cases}$$